A photograph taken from the International Space Station (ISS) showing the Earth's surface. The aurora borealis is visible as a bright green glow in the upper atmosphere. The ISS structure, including solar panel arrays, is visible in the foreground.

# Surface Charging: Validation Challenges and Suggestions

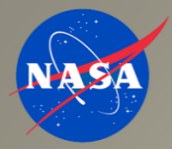
**Joseph Minow**

**NASA, Langley Research Center**

**9<sup>th</sup> Community Coordinated Modeling Center Workshop 2018**

**College Park, Maryland 23-27 April 2018**

**[joseph.minow@nasa.gov](mailto:joseph.minow@nasa.gov)**



# Surface Charging Challenges

- Surface charging physics is a current balance phenomenon

$$\frac{dQ}{dt} = C \frac{dV}{dt} = \frac{d\sigma}{dt} A = \sum_k I_k$$

- Surface charging threat environments

Environment	Extreme Potential
LEO auroral	-1 to -2 kV
LEO solar array	~90% of string voltage
GEO (GTO)	-10 kV to -20 kV

$$\frac{dQ}{dt} = \sum_k I_k =$$

- + I<sub>i</sub> (V)      **incident ions**
- I<sub>e</sub> (V)      **incident electrons**
- + I<sub>bs,e</sub> (V)    **backscattered electrons**
- ± I<sub>c</sub> (V)      **conduction currents**
- + I<sub>se</sub> (V)      **secondary electrons due to I<sub>e</sub>**
- + I<sub>si</sub> (V)      **secondary electrons due to I<sub>i</sub>**
- + I<sub>ph,e</sub> (V)    **photoelectrons**

- Validating that a space weather model and charging code can be used to predict surface charging requires validating two models:
  - Does the space weather model correctly generate the relevant environments?
  - Does a spacecraft charging code generate the correct surface potential (or electric field) using space weather environment model inputs?

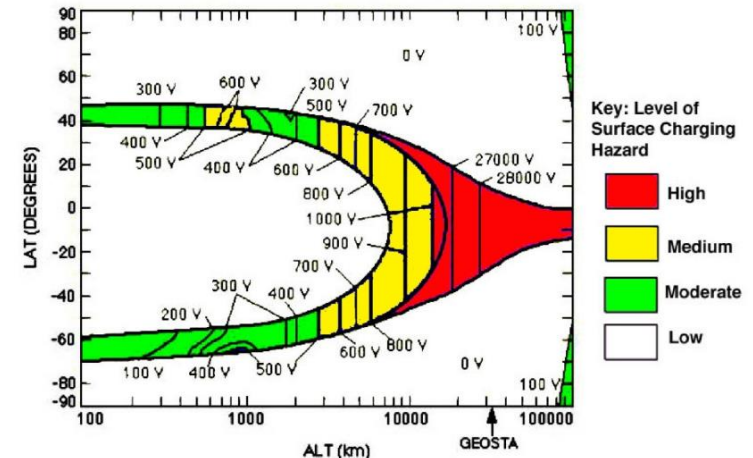
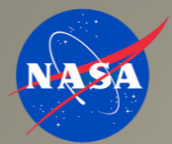
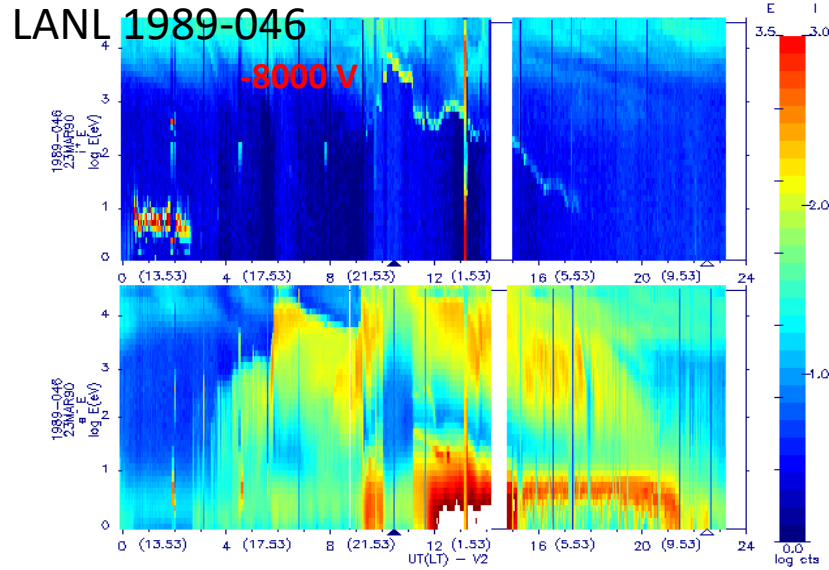


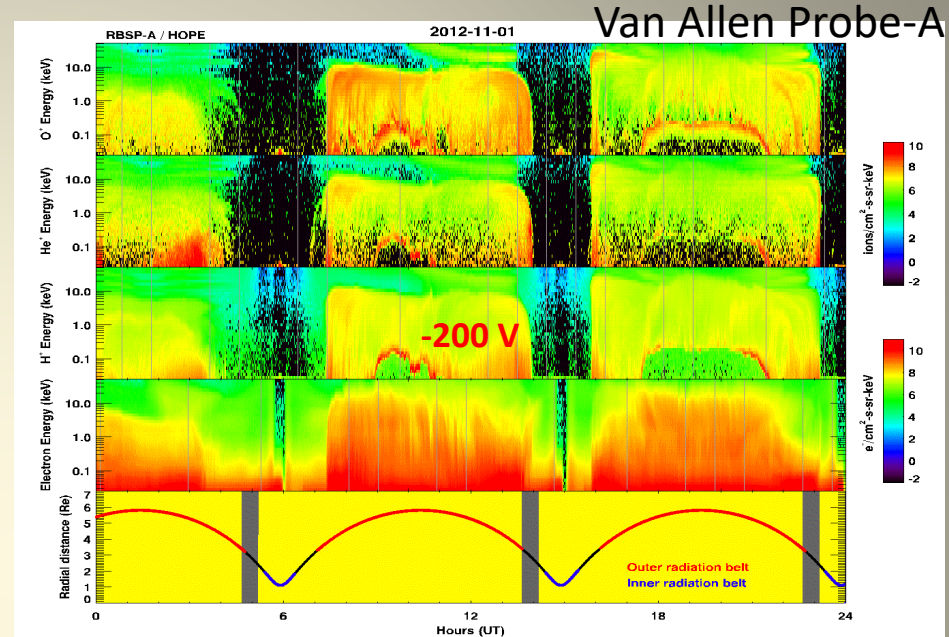
Figure 1—Earth Regimes of Concern for On-Orbit Surface Charging Hazards for Spacecraft Passing Through Indicated Latitude and Altitude (Evans and others (1989))



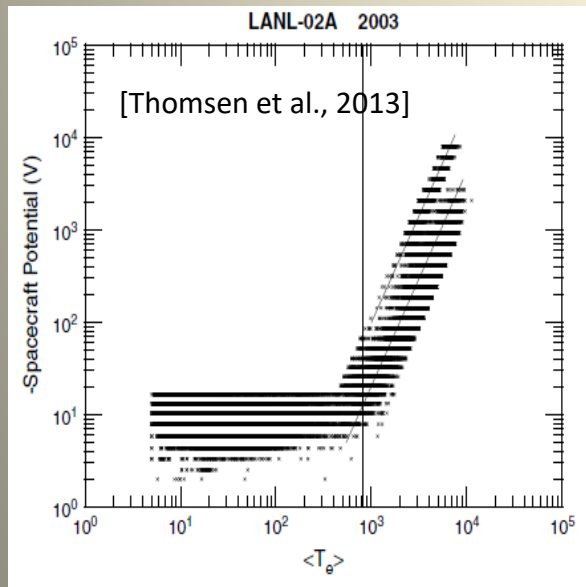
# GEO (Outer Radiation Belt) Charging



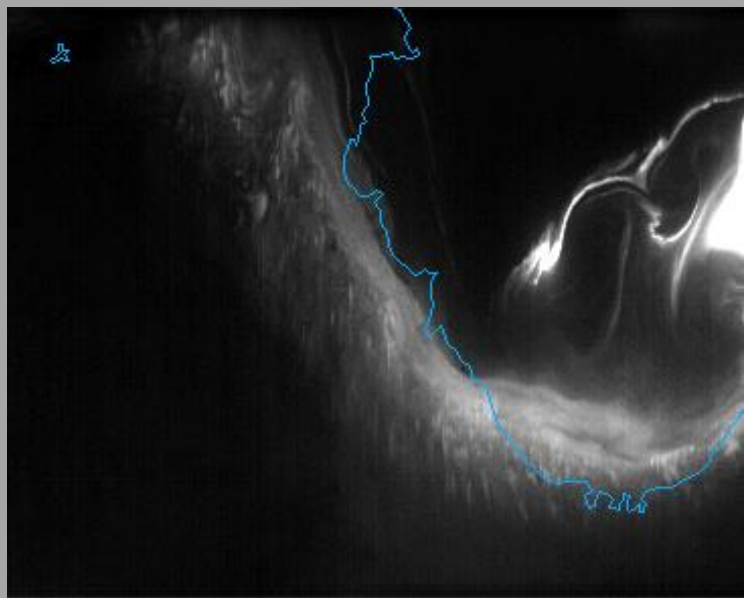
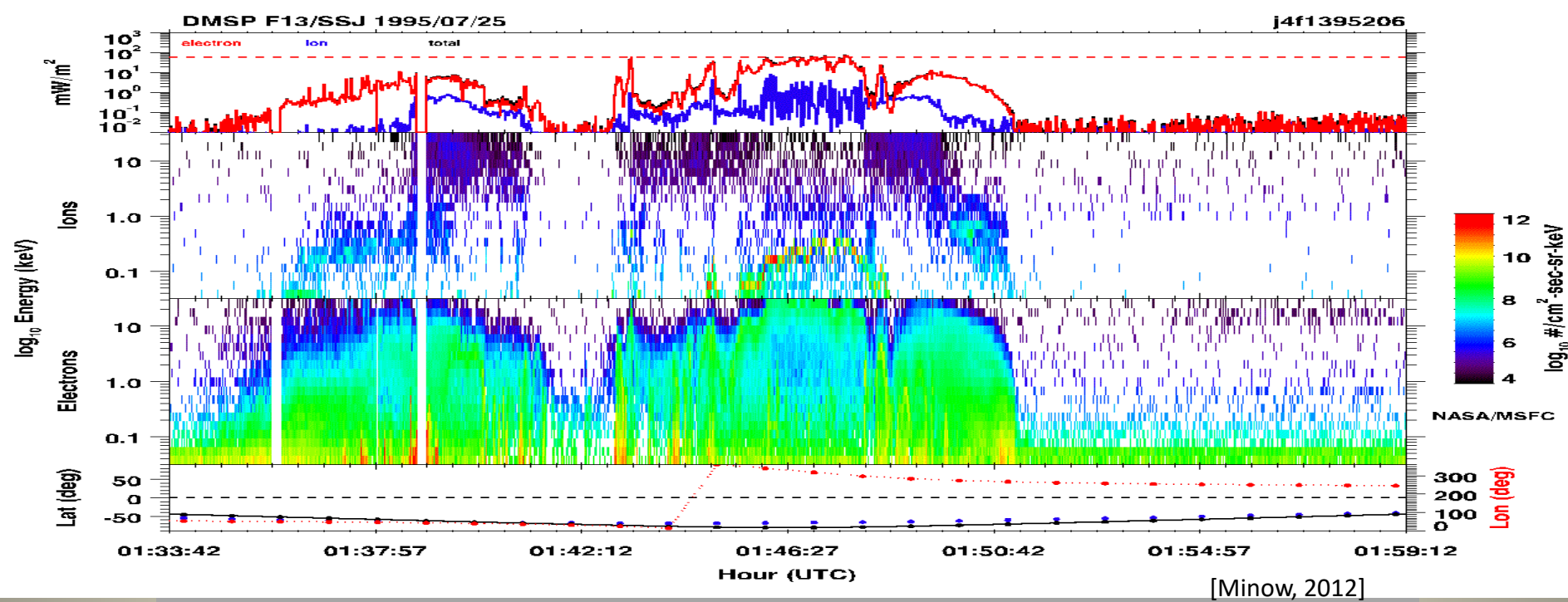
[<https://www.mpa.lanl.gov>]



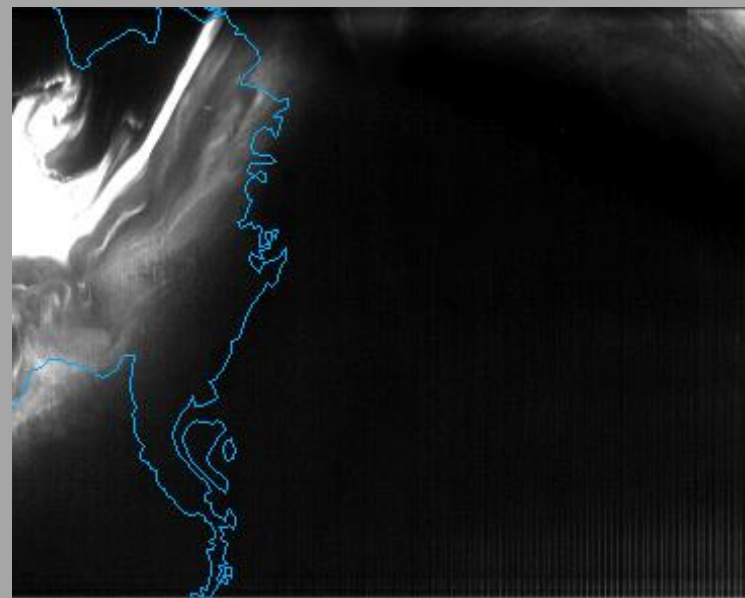
[Parker and Minow, 2014]



- GEO charging is driven by the hot thermal plasma during storm environments
- Space weather models need to predict the background
- Charging code inputs are typically Ne, Te parameters for a single or bi-maxwellian fits to electron and ion flux

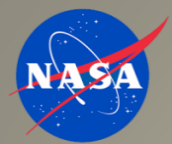


1995-07-25 01:33:42.0



1995-07-25 01:46:27.0

1995-07-25 01:59:12.0



# Fontheim Distribution

## Ambient background

$n=1.0e10$        $1/m^3$   
 $T_e=0.2$       eV

## Maxwellian

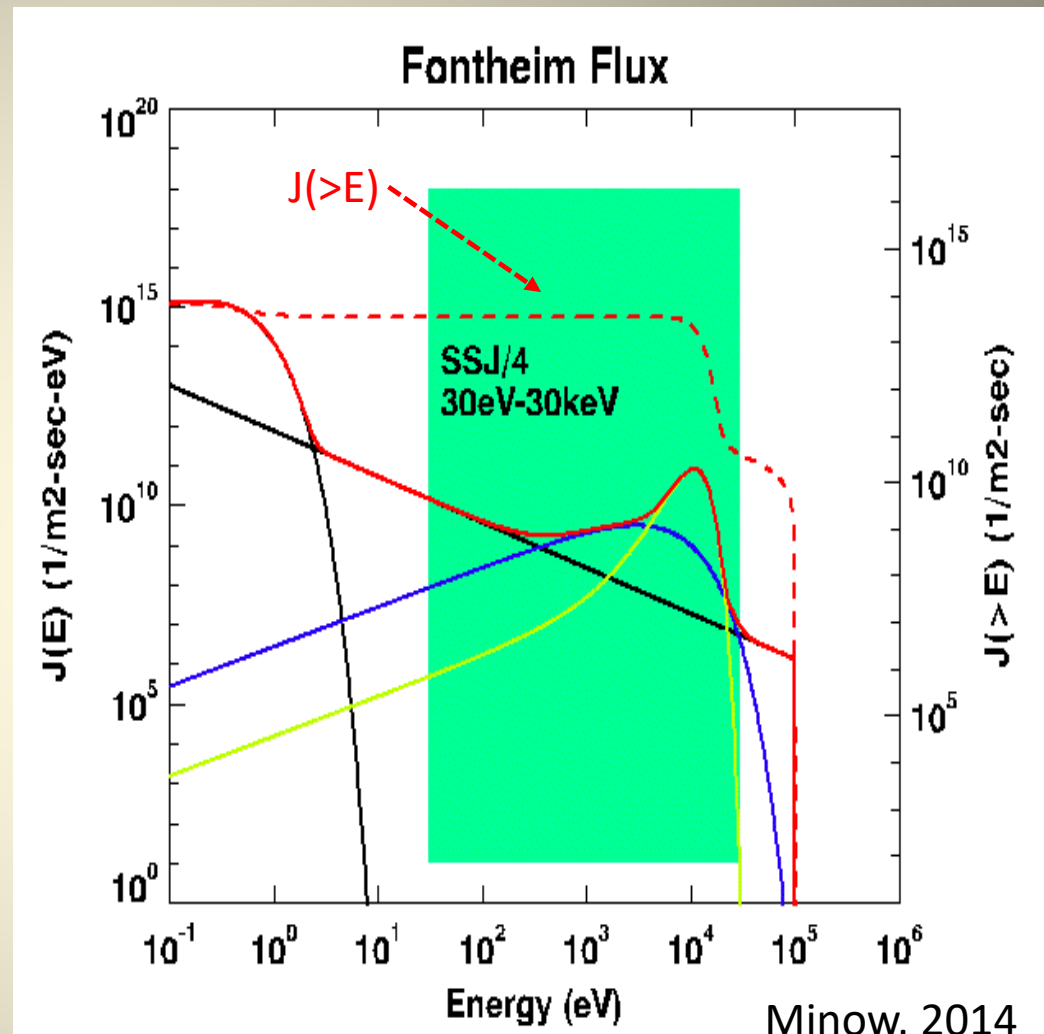
$J_{max} = 4.0e-6$        $A/m^2$   
 $T_e = 3.0e3$       eV

## Gaussian (beam)

$J_{gau} = 0.9e-4$        $A/m^2$   
 $E_{gau} = 10.0e3$       eV beam energy  
 $d_{gau} = 4.0e3$       eV beam width

## Power Law

$J_{pwr} = 3.0e-7$        $A/m^2$   
 $\alpha = 1.15$       exponent  
 $E_1=50.0$       eV, first energy  
 $E_2=1.0e5$       eV, second energy



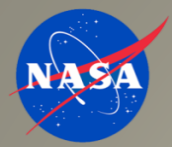
$$\text{Flux}(E) = \sqrt{\frac{e}{2\pi\theta m_e}} \frac{E}{\theta} n \exp\left(-\frac{E}{\theta}\right) + \pi\zeta_{\max} E \exp\left(-\frac{E}{\theta_{\max}}\right) + \pi\zeta_{\text{gauss}} E \exp\left(-\left(\frac{E_{\text{gauss}} - E}{\Delta}\right)^2\right) + \pi\zeta_{\text{power}} E^{-\alpha}$$



# Surface Charging Suggestions\*

---

- Initial effort will focus on high priority GEO, MEO, GTO, and LEO polar environments where surface charging can exceed hundreds of volts
- User groups include spacecraft designers, operational situational awareness, anomaly investigations, and impact on science measurements
- Metrics (team is evaluating options):
  - Statistical evaluation using O'Brien "green anomalies" technique
  - Parameters used for inputs to charging models
    - GEO, MEO, GTO: Ne, Te, Ni, Ti or other
    - LEO polar (auroral): Ne,  $E_{\text{beam}}$ ,  $\Delta E_{\text{beam}}$ , and other Fontheim parameters
    - Flux spectra at different locations
- Environment models (initial focus):
  - Ovation – CCMC implementation
  - LANL model (Vania Jordanova)
  - IMPTAM (Natalia Ganjushkina), run online in near-real time since 2013
  - CIMI (Natalia Buzulukova)
- Spacecraft charging models (secondary effort, but compare with  $\Phi_{s/c}$ )
  - Nascap
  - SPIS
  - SPENVIS, MUSCAT, and other small group charging codes



# Metrics (I): Surface Charging

---

- The traditional ones
  - RMSE root mean square error
  - Ratio of maximum amplitude(flux)
  - Prediction efficiency
  
- Above a certain threshold (yes/no prediction)
  - $>10$  keV flux exceeds  $1.5e7$  ( $1/cm^2/sec/ster$ ) (surface charging threshold)
  - Whether a model prediction predicts a surface charging event during a certain interval with the defined threshold
  - Generation of a contingency table of hit (H), miss (M), false positive (F), and correct negative (N) model predictions
  - Heidke Skill Score (for a large enough surface charging events)



# Metrics (II): Surface Charging

---

- New ones using log ratio (Morley et al., 2018) satisfying the four attributes (see below)

The median symmetric accuracy

$$\zeta = 100(\exp(M(|\log_e(Q_i)|)) - 1)$$

where  $Q_i = \frac{y_i}{x_i}$ : ratio of predicted versus observed,  $y_i$ : model,  $x_i$ : observation

M: median value

The Symmetric Signed Percentage Bias (SSPB):

$$SSPB = 100 \operatorname{sgn} \left( M(\log_e(Q_i)) \right) (\exp(|M(\log_e(Q_i))|) - 1)$$

Sgn: signum function; M: median value

- (1) The metrics must be meaningful for data that cover orders of magnitude,
- (2) underprediction and overprediction by the same factor should be penalized equally,
- (3) the metrics should be easy to interpret, and
- (4) the metrics should be robust to the presence of outliers and bad data.





Questions?